

## Thermodynamic Design and Aerodynamic Analysis of Supercritical Carbon Dioxide Turbine

Dongbo Shi<sup>1,a</sup>, Liangliang Li<sup>1,b</sup>, Yanyao Zhang<sup>1,c</sup> and Yonghui Xie<sup>1,d</sup>

<sup>1</sup>School of Energy and Power Engineering, Xi'an Jiaotong University, Xi'an, China

<sup>a</sup>sdb\_xjtu@126.com, <sup>b</sup>liliangliang\_xjtu@163.com, <sup>c</sup>zhangyanyao@stu.xjtu.edu.cn, <sup>d</sup>yhxie@mail.xjtu.edu.cn

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**Abstract.** Supercritical carbon dioxide power cycle can reduce the power consumption and achieve higher system efficiency with the properties mutation phenomenon of supercritical fluids near the critical point. Turbine is the critical component of the heat-work conversion in cycle which has advantages of compact structure, small size and so on. It has significant engineering values for its design. In this study, based on one dimensional flow, the RNG k-epsilon turbulence model was used. The supercritical carbon dioxide turbine nozzle and impeller were designed preliminarily, and the three-dimensional flow model was established. The detailed parameters of the aerodynamic performance were obtained. The results are expected to provide the appropriate design scheme and reference data for the engineering application of the supercritical carbon dioxide turbine and power cycling system.

### Introduction

Compared with the axial turbine, radial turbine has the advantages of compact structure, simple manufacturing process, low cost, and high efficiency at the design conditions of small flow, etc. Therefore, in recent years with the rapid development of engineering and technology, radial turbine is used more and more which had been overlooked over a period of time. The research of radial turbine at home and abroad is more concentrated in the conventional working fluid like gas or steam [1,2]. Supercritical carbon dioxide is a kind of supercritical fluid, its critical conditions are easy to achieve which the critical temperature is 31.1°C and the critical pressure is 7.4MPa [3]. Furthermore, it's an environmentally friendly natural working fluid. In this paper, the thermodynamic design and aerodynamic characteristics of supercritical carbon dioxide turbine are obtained based on one dimensional flow.

### Thermodynamic Design

Based on one dimensional flow with the real gas state equation, flow continuous equation, momentum equation and energy equation, the characteristic thermodynamic parameters of turbine nozzle and impeller are obtained by advance calculation, scheme comparison and actual calculation which are shown in Table 1. And ultimately the geometric structure dimensions of turbine could be determined. The type line of the nozzle and the shape of the impeller in meridian plane are shown in Fig.1 and Fig.2.

Table 1 The Characteristic Thermodynamic Parameters

Projects	Value	Projects	Value
Amount of expansion gas [kg/s]	8.646	The efficiency of the flow channel	89.75%
Inlet pressure [MPa]	20	Efficiency	84.17%
Inlet temperature [K]	373.15	Shaft power [kW]	198.06
Outlet pressure [Mpa]	7.8	Rotating speed [rad/s]	12504.49

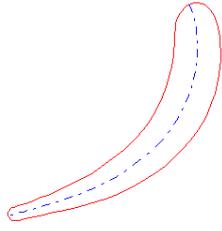


Fig.1 The type line of the nozzle

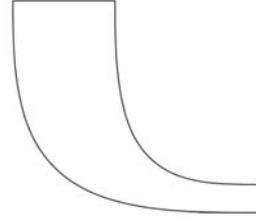


Fig.2 The shape of the impeller in meridian plane

### Physical Model

The structure of the computing area and the mesh of the single nozzle and impeller are shown in Fig.3. In the calculation, the hexahedral structured mesh is used, and a mesh size of 256156 is used in the single nozzle while 266178 used in the single impeller.

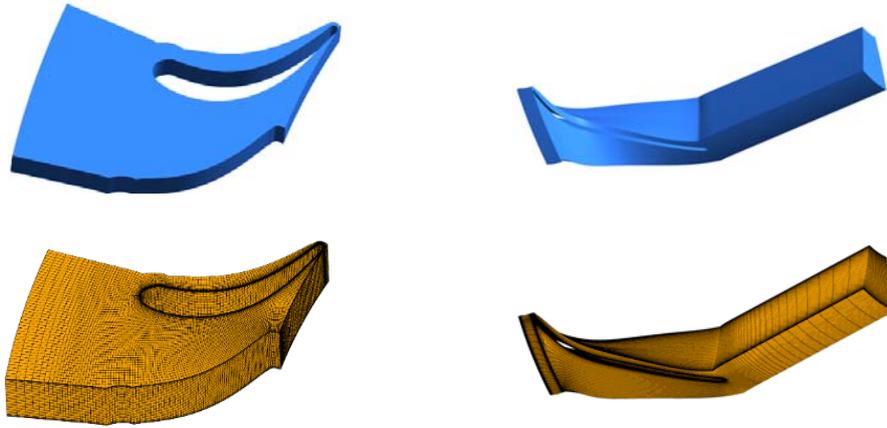


Fig.3 The structure of the computing area and the mesh of the single nozzle and impeller

### Results and Discussions

**Results for the Nozzle.** The velocity distribution and an enlarged view of full circle nozzles are shown in Fig.4. The windward stagnation zone and the stagnation point at the leading edge of the nozzle blade can be observed clearly. The velocity gradient is large at about 60% relative chord place and the maximum velocity is increased to approximately 468m/s. It should be noted that the fluid velocity at the nozzle outlet is slightly decreased, but the overall distribution is more uniform. And it maintains a higher velocity value.

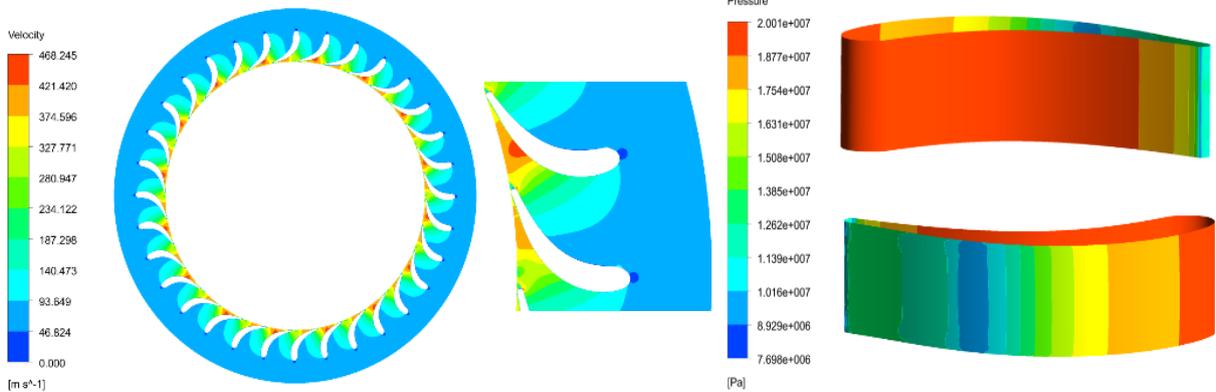


Fig.4 The velocity distribution of the nozzle

Fig.5 The pressure distribution of the nozzle blade

The pressure distribution of the single nozzle is shown in Fig.5. At the pressure side, the pressure is large and remained unchanged from the leading edge to about 85% relative chord place. From the

followed position to the trailing edge, the pressure gradient is large and the lowest pressure is about 10MPa located at the nozzle outlet. The pressure distribution is completely different at the suction side. The greater pressure gradient is appeared near the leading edge and the pressure is reduced to about 9MPa at about 60% relative chord place. With the flow towards the downstream, the pressure of the suction surface rises slowly at first, and then it is decreased slightly at the outlet.

**Results for the Impeller.** Fig.6 shows the absolute velocity distribution of the sections at the 10%, 50% and 90% blade height respectively and partially enlarged views. It can be seen that the smaller velocity appears near the leading edge of the pressure side because of the vortex and the maximum velocity appears near the leading edge of the suction side. Close to the suction side, the flow is relatively steady and the velocity changes smoothly. With the increase of the blade height the flow is more relaxed.

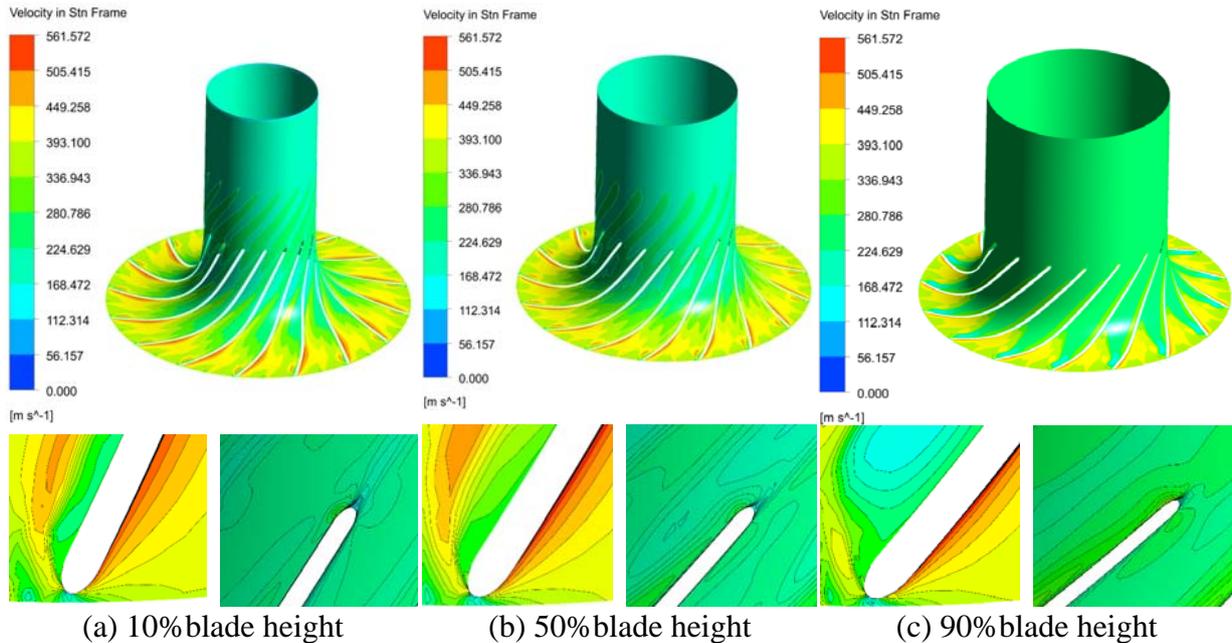


Fig.6 The absolute velocity distribution of the impeller

The pressure distribution of the impeller blade is shown in Fig.7. At the pressure side, the pressure gradually decreases along flowing direction. At the suction side, from the leading edge to about 35% of the blade chord area, the pressure is gradually reduced. Pressure is increased and basically unchanged from the followed position to the trailing edge. At the same time, it should be noticed that there is a significant pressure gradient along the direction of the blade height in about 50% of the chord to the trailing edge region. This is due to the distortion of the blades and the growth of the blade height.

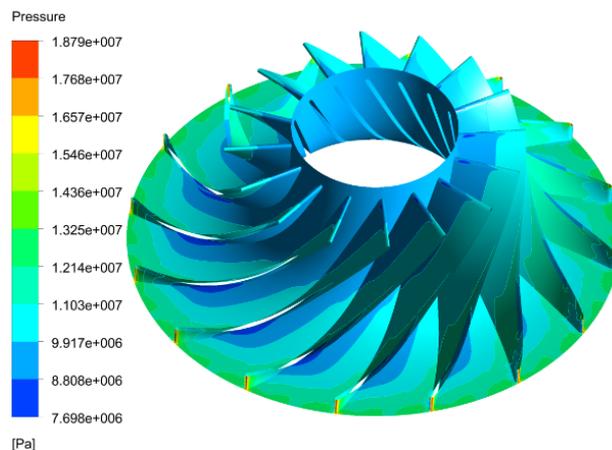


Fig.7 The pressure distribution of the impeller blade

**Results for Power and Efficiency.** Table 2 shows the results comparison of numerical simulation and thermodynamic calculation. It shows that the power and efficiency of numerical simulation are lower, because of the supersonic flow in the turbine, the shock loss is produced.

Table 2 Results Comparison

Projects	Thermodynamic calculation	Numerical simulation
Amount of expansion gas [kg/s]	2.276	2.8788
Outlet temperature [K]	744.12	747.697
Outlet enthalpy [J/kg]	948081.32	952350
Shaft power [kW]	198.056	187.00
Efficiency	0.8417	0.78499

## Conclusion

In this paper, the thermodynamic design and the detailed parameters of the aerodynamic performance of the supercritical carbon dioxide turbine are obtained. The main conclusions are as follows:

- (1) Through the thermodynamic design of the turbine, the thermodynamic parameters of each feature point and the basic dimensions of main parts are obtained. The power of the designed turbine is 198.056kW and the efficiency is 84.17%.
- (2) The velocity and pressure distribution are obtained while the RNG  $k-\varepsilon$  turbulence model is used in the numerical simulation.
- (3) After numerical calculation, the power of the turbine is 187.00kW and the efficiency is 78.50%. Because of the supersonic flow in the turbine, the power and efficiency of numerical simulation are lower.

## References

- [1] Wasserbauer C A, Glassman A: T.FORTRAN Programming for Predicting Off-Design Performance of Radial In-flow Turbines, NASA Technical Report TN D-8063, 1975
- [2] QiuXW, N Baines: Performance Prediction for High Pressure-Ratio Radial Inflow Turbines, proceedings of the ASME Turbo Expo.2007:945-956, 1868.
- [3] V. Dostal, M.J. Driscoll, P. Hejzlar: A Supercritical Carbon Dioxide Cycle for Next Generation Nuclear Reactors, (2004).