

(d) Z direction stress curve with engine speed

Fig.4.2 Node 3370 of the stress response curve with engine speed

From the figure.4.1 and figure.4.2 can be seen in the existing compressor blade convex shoulder structure dry friction damping effect is very obvious, in the whole speed range of each direction vibration displacement and stress has a lot of lower. Displacement maximum response by coordination of 2.75 mm (4700 rpm) be reduced to consider dry friction damping effect of 0.43 mm (4800 rpm); The maximum equivalent stress by coordination of 184 Mpa (4700 rpm) reduced to consider dry friction damping effect 24 Mpa (4800 rpm).

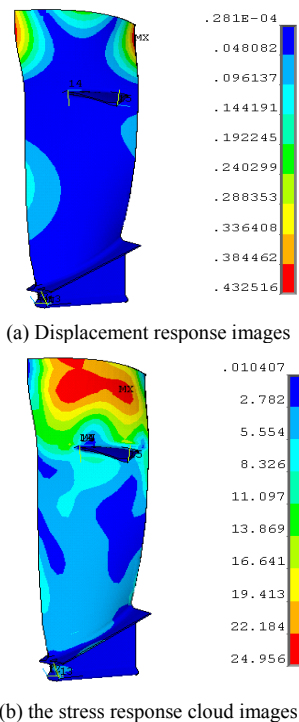


Fig.4.3 Consider the maximum response of the fan blades cloud images in the operating speed range of design state dry friction damping

Calculate the stress response for different frequency blade under stress and displacement response of the distribution by finite element method. Consider dry friction damping action get work speed within the scope of the

maximum response vibration mode, and don't take time two node displacements coordinate the entire sixth order resonance vibration mode similar, but the response amplitude greatly reduced. Maximum stress response distribution and does not consider the dry friction stress distribution is also similar, but stress amplitude greatly reduced.

V. CONCLUSIONS

1) Different friction coefficient corresponding tangential contact stiffness with positive pressure change curve. Contact surface tangential stiffness with normal positive pressure increases while the, but the increase rate tends to slow. Friction coefficients on the tangential stiffness influence is mainly embodied in the positive pressure is less than 400 n, when positive pressure is large friction coefficient of the response is very small.

2) In this paper research compressor blade speaking. Through calculation, the actual engine work process contact only area of contact and pressure distribution is not uniform. So according to the above analysis, we can know, for the compressor blade speaking, the contact stiffness value influenced by the contact surface shape and the influence of the friction coefficient is very small, and working process of contact stiffness should be changed little.

3) Convex shoulder structure of different contact surface state will not only affect the blade resonance peak size, also will change resonance speed. Therefore, when the design to comprehensive consideration of the damping effect and resonance speed, best can directly to avoid harmful resonance; If avoided, should make vibration stress reduced to a reasonable level.

REFERENCES

- [1] Weiqiang Zhao, Yongxian Liu and Mowu Lu, "FEA on vibration characteristics analysis of an aero-engine compressor blade," IEEE Control Syst.Soc. China, vol. C7440, pp. 3248-50, May 2012.
- [2] Bing Chen, Ting Yang and Tao Feng, "Ontology-based aero-engine compressor blade manufacturing quality analyzing," IEEE Wuhan Sect. China, vol C7480, pp.4, Dec 2009.
- [3] Hammouda, M.M.I and Pasha, R.A, "Modelling of cracking sites/development in axial dovetail joints of aero-engine compressor discs," International Journal of Fatigue. Vol 29, pp.30-48, January 2007.
- [4] Weiqiang Zhao, Yongxian Liu and Mowu Lu, "Vibration analysis of aero-engine compressor blade-disc coupling system," Appl.Mech.Mater. Vol 16-19, pp.264-268, 2009.
- [5] Savkov,K, Doroshko,S and Urbaha,M, "Gas turbine aero engine compressor blade erosion and heat resistant coatings," Transp.Means-Proc.Int.Conf. pp.274-278, October 2011.
- [6] Wei-Feng He, Ying-Hong Li and Wei Li, "Laser shock peening on vibration fatigue behavior of compressor blade," Hongkong Dongli Xuebao. Vol 26, pp.1551-1556, July 2011.